

Start testing: Feb 17 12:24 W. Europe Standard Time

1/137 Testing: test_AerodynamicMomentAndAerodynamicForce

1/137 Test: test_AerodynamicMomentAndAerodynamicForce

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_AerodynamicMomentAndAerodynamicForce.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Aerodynamics

"test_AerodynamicMomentAndAerodynamicForce" start time: Feb 17 12:24 W. Europe Standard Time

Output:

Running 3 test cases...

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle
21, Rotation matrix from inertial frame to corotating frame of Vehicle
30, Rotation matrix from inertial frame to trajectory frame of Vehicle
39, Airspeed-based velocity of Vehicle
42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth
3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle
12, Rotation matrix from inertial frame to body frame of Vehicle
21, Rotation matrix from inertial frame to corotating frame of Vehicle
30, Rotation matrix from inertial frame to trajectory frame of Vehicle
39, Airspeed-based velocity of Vehicle
42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth
3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle
12, Rotation matrix from inertial frame to body frame of Vehicle
21, Rotation matrix from inertial frame to corotating frame of Vehicle
30, Rotation matrix from inertial frame to trajectory frame of Vehicle
39, Airspeed-based velocity of Vehicle
42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth
3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle
12, Rotation matrix from inertial frame to body frame of Vehicle
21, Rotation matrix from inertial frame to corotating frame of Vehicle
30, Rotation matrix from inertial frame to trajectory frame of Vehicle
39, Airspeed-based velocity of Vehicle
42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle
21, Rotation matrix from inertial frame to corotating frame of Vehicle
30, Rotation matrix from inertial frame to trajectory frame of Vehicle
39, Airspeed-based velocity of Vehicle
42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth
3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle
12, Rotation matrix from inertial frame to body frame of Vehicle
21, Rotation matrix from inertial frame to corotating frame of Vehicle
30, Rotation matrix from inertial frame to trajectory frame of Vehicle
39, Airspeed-based velocity of Vehicle
42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth
3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle
12, Rotation matrix from inertial frame to body frame of Vehicle
21, Rotation matrix from inertial frame to corotating frame of Vehicle
30, Rotation matrix from inertial frame to trajectory frame of Vehicle
39, Airspeed-based velocity of Vehicle
42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth
3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle
12, Rotation matrix from inertial frame to body frame of Vehicle
21, Rotation matrix from inertial frame to corotating frame of Vehicle
30, Rotation matrix from inertial frame to trajectory frame of Vehicle
39, Airspeed-based velocity of Vehicle
42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type aerodynamic , acting on Vehicle, exerted by Earth

3, Rotation matrix from inertial frame to aerodynamic frame of Vehicle

12, Rotation matrix from inertial frame to body frame of Vehicle

21, Rotation matrix from inertial frame to corotating frame of Vehicle

30, Rotation matrix from inertial frame to trajectory frame of Vehicle

39, Airspeed-based velocity of Vehicle

42, Single acceleration in inertial frame of type thrust , acting on Vehicle

*** No errors detected

<end of output>

Test time = 3.98 sec

Test Passed.

"test_AerodynamicMomentAndAerodynamicForce" end time: Feb 17 12:24 W. Europe Standard Time

"test_AerodynamicMomentAndAerodynamicForce" time elapsed: 00:00:03

2/137 Testing: test_AerodynamicsNamespace

2/137 Test: test_AerodynamicsNamespace

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_AerodynamicsNamespace.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Aerodynamics

"test_AerodynamicsNamespace" start time: Feb 17 12:24 W. Europe Standard Time

Output:

Running 2 test cases...

*** No errors detected

<end of output>

Test time = 2.00 sec

Test Passed.

"test_AerodynamicsNamespace" end time: Feb 17 12:24 W. Europe Standard Time

"test_AerodynamicsNamespace" time elapsed: 00:00:02

3/137 Testing: test_AerodynamicCoefficientGenerator

3/137 Test: test_AerodynamicCoefficientGenerator

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_AerodynamicCoefficientGenerator.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Aerodynamics

"test_AerodynamicCoefficientGenerator" start time: Feb 17 12:24 W. Europe Standard Time

Output:

Running 2 test cases...

*** No errors detected

<end of output>

Test time = 2.14 sec

Test Passed.

"test_AerodynamicCoefficientGenerator" end time: Feb 17 12:24 W. Europe Standard Time

"test_AerodynamicCoefficientGenerator" time elapsed: 00:00:02

4/137 Testing: test_ExponentialAtmosphere

4/137 Test: test_ExponentialAtmosphere

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_ExponentialAtmosphere.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Aerodynamics

"test_ExponentialAtmosphere" start time: Feb 17 12:24 W. Europe Standard Time

Output:

Running 4 test cases...

*** No errors detected

<end of output>

Test time = 2.11 sec

Test Passed.

"test_ExponentialAtmosphere" end time: Feb 17 12:24 W. Europe Standard Time

"test_ExponentialAtmosphere" time elapsed: 00:00:02

5/137 Testing: test_TabulatedAtmosphere

5/137 Test: test_TabulatedAtmosphere

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_TabulatedAtmosphere.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Aerodynamics

"test_TabulatedAtmosphere" start time: Feb 17 12:24 W. Europe Standard Time

Output:

Running 5 test cases...

*** No errors detected

<end of output>

Test time = 2.15 sec

Test Passed.

"test_TabulatedAtmosphere" end time: Feb 17 12:24 W. Europe Standard Time

"test_TabulatedAtmosphere" time elapsed: 00:00:02

6/137 Testing: test_TabulatedAerodynamicCoefficients

6/137 Test: test_TabulatedAerodynamicCoefficients

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_TabulatedAerodynamicCoefficients.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Aerodynamics

"test_TabulatedAerodynamicCoefficients" start time: Feb 17 12:24 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 5.50 sec

Test Passed.

"test_TabulatedAerodynamicCoefficients" end time: Feb 17 12:24 W. Europe Standard Time

"test_TabulatedAerodynamicCoefficients" time elapsed: 00:00:05

7/137 Testing: test_HeatTransfer

7/137 Test: test_HeatTransfer

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_HeatTransfer.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Aerodynamics

"test_HeatTransfer" start time: Feb 17 12:24 W. Europe Standard Time

Output:

Running 3 test cases...

*** No errors detected

<end of output>

Test time = 1.52 sec

Test Passed.

"test_HeatTransfer" end time: Feb 17 12:24 W. Europe Standard Time

"test_HeatTransfer" time elapsed: 00:00:01

8/137 Testing: test_ControlSurfaceIncrements

8/137 Test: test_ControlSurfaceIncrements

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_ControlSurfaceIncrements.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Aerodynamics

"test_ControlSurfaceIncrements" start time: Feb 17 12:24 W. Europe Standard Time

Output:

Running 2 test cases...

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Mach number of Apollo

1, Body orientation angle angle of attack of Apollo

2, Body orientation angle sideslip angle of Apollo

3, Control Surface Deflection of Apollo w.r.t. TestSurface

4, Aerodynamic moment coefficients of Apollo

7, Aerodynamic force coefficients of Apollo

*** No errors detected

<end of output>

Test time = 3.40 sec

Test Passed.

"test_ControlSurfaceIncrements" end time: Feb 17 12:24 W. Europe Standard Time

"test_ControlSurfaceIncrements" time elapsed: 00:00:03

9/137 Testing: test_AerodynamicCoefficientsFromFile

9/137 Test: test_AerodynamicCoefficientsFromFile

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_AerodynamicCoefficientsFromFile.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Aerodynamics

"test_AerodynamicCoefficientsFromFile" start time: Feb 17 12:24 W. Europe Standard Time

Output:

Running 1 test case...

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Mach number of SpacePlane

1, Body orientation angle angle of attack of SpacePlane

2, Body orientation angle sideslip angle of SpacePlane

3, Body orientation angle bank angle of SpacePlane

4, Aerodynamic moment coefficients of SpacePlane

7, Aerodynamic force coefficients of SpacePlane

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Mach number of SpacePlane

1, Body orientation angle angle of attack of SpacePlane

2, Body orientation angle sideslip angle of SpacePlane

3, Body orientation angle bank angle of SpacePlane

4, Control Surface Deflection of SpacePlane w.r.t. TestSurface

5, Aerodynamic moment coefficients of SpacePlane

8, Aerodynamic force coefficients of SpacePlane

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Mach number of SpacePlane

1, Body orientation angle angle of attack of SpacePlane

2, Body orientation angle sideslip angle of SpacePlane

3, Body orientation angle bank angle of SpacePlane

4, Control Surface Deflection of SpacePlane w.r.t. TestSurface

5, Aerodynamic moment coefficients of SpacePlane

8, Aerodynamic force coefficients of SpacePlane

*** No errors detected

<end of output>

Test time = 2.60 sec

Test Passed.

"test_AerodynamicCoefficientsFromFile" end time: Feb 17 12:24 W. Europe Standard Time

"test_AerodynamicCoefficientsFromFile" time elapsed: 00:00:02

10/137 Testing: test_NRLMSISE00Atmosphere

10/137 Test: test_NRLMSISE00Atmosphere

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_NRLMSISE00Atmosphere.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Aerodynamics

"test_NRLMSISE00Atmosphere" start time: Feb 17 12:24 W. Europe Standard Time

Output:

Running 24 test cases...

*** No errors detected

<end of output>

Test time = 2.30 sec

Test Passed.

"test_NRLMSISE00Atmosphere" end time: Feb 17 12:24 W. Europe Standard Time

"test_NRLMSISE00Atmosphere" time elapsed: 00:00:02

11/137 Testing: test_AstrodynamicsFunctions

11/137 Test: test_AstrodynamicsFunctions

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_AstrodynamicsFunctions.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/BasicAstrodynamics

"test_AstrodynamicsFunctions" start time: Feb 17 12:24 W. Europe Standard Time

Output:

Running 5 test cases...

*** No errors detected

<end of output>

Test time = 1.82 sec

Test Passed.

"test_AstrodynamicsFunctions" end time: Feb 17 12:24 W. Europe Standard Time

"test_AstrodynamicsFunctions" time elapsed: 00:00:01

12/137 Testing: test_OrbitalElementConversions

12/137 Test: test_OrbitalElementConversions

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_OrbitalElementConversions.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/BasicAstrodynamics

"test_OrbitalElementConversions" start time: Feb 17 12:24 W. Europe Standard Time

Output:

Running 11 test cases...

*** No errors detected

<end of output>

Test time = 1.47 sec

Test Passed.

"test_OrbitalElementConversions" end time: Feb 17 12:24 W. Europe Standard Time

"test_OrbitalElementConversions" time elapsed: 00:00:01

13/137 Testing: test_PhysicalConstants

13/137 Test: test_PhysicalConstants

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_PhysicalConstants.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/BasicAstrodynamics

"test_PhysicalConstants" start time: Feb 17 12:24 W. Europe Standard Time

Output:

Running 4 test cases...

*** No errors detected

<end of output>

Test time = 1.47 sec

Test Passed.

"test_PhysicalConstants" end time: Feb 17 12:24 W. Europe Standard Time

"test_PhysicalConstants" time elapsed: 00:00:01

14/137 Testing: test_UnitConversions

14/137 Test: test_UnitConversions

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_UnitConversions.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/BasicAstrodynamics

"test_UnitConversions" start time: Feb 17 12:24 W. Europe Standard Time

Output:

Running 14 test cases...

*** No errors detected

<end of output>

Test time = 1.86 sec

Test Passed.

"test_UnitConversions" end time: Feb 17 12:25 W. Europe Standard Time

"test_UnitConversions" time elapsed: 00:00:01

15/137 Testing: test_MeanToEccentricAnomalyConversion

15/137 Test: test_MeanToEccentricAnomalyConversion

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_MeanToEccentricAnomalyConversion.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/BasicAstrodynamics

"test_MeanToEccentricAnomalyConversion" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 7 test cases...

*** No errors detected

<end of output>

Test time = 4.13 sec

Test Passed.

"test_MeanToEccentricAnomalyConversion" end time: Feb 17 12:25 W. Europe Standard Time

"test_MeanToEccentricAnomalyConversion" time elapsed: 00:00:04

16/137 Testing: test_MeanToHyperbolicEccentricAnomalyConversion

16/137 Test: test_MeanToHyperbolicEccentricAnomalyConversion

Command:

"C:/tudatBundle/tudat/bin/unit_tests/test_MeanToHyperbolicEccentricAnomalyConversion.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/BasicAstrodynamics

"test_MeanToHyperbolicEccentricAnomalyConversion" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 5 test cases...

*** No errors detected

<end of output>

Test time = 5.86 sec

Test Passed.

"test_MeanToHyperbolicEccentricAnomalyConversion" end time: Feb 17 12:25 W. Europe Standard Time

"test_MeanToHyperbolicEccentricAnomalyConversion" time elapsed: 00:00:05

17/137 Testing: test_KeplerPropagator

17/137 Test: test_KeplerPropagator

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_KeplerPropagator.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/BasicAstrodynamics

"test_KeplerPropagator" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 6 test cases...

*** No errors detected

<end of output>

Test time = 1.31 sec

Test Passed.

"test_KeplerPropagator" end time: Feb 17 12:25 W. Europe Standard Time

"test_KeplerPropagator" time elapsed: 00:00:01

18/137 Testing: test_AccelerationModel

18/137 Test: test_AccelerationModel

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_AccelerationModel.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/BasicAstrodynamics

"test_AccelerationModel" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 3 test cases...

*** No errors detected

<end of output>

Test time = 1.48 sec

Test Passed.

"test_AccelerationModel" end time: Feb 17 12:25 W. Europe Standard Time

"test_AccelerationModel" time elapsed: 00:00:01

19/137 Testing: test_ClohessyWiltshirePropagator

19/137 Test: test_ClohessyWiltshirePropagator

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_ClohessyWiltshirePropagator.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/BasicAstrodynamics

"test_ClohessyWiltshirePropagator" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 2 test cases...

*** No errors detected

<end of output>

Test time = 1.47 sec

Test Passed.

"test_ClohessyWiltshirePropagator" end time: Feb 17 12:25 W. Europe Standard Time

"test_ClohessyWiltshirePropagator" time elapsed: 00:00:01

20/137 Testing: test_MissionGeometry

20/137 Test: test_MissionGeometry

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_MissionGeometry.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/BasicAstrodynamics

"test_MissionGeometry" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 6 test cases...

*** No errors detected

<end of output>

Test time = 1.69 sec

Test Passed.

"test_MissionGeometry" end time: Feb 17 12:25 W. Europe Standard Time

"test_MissionGeometry" time elapsed: 00:00:01

21/137 Testing: test_TimeConversions

21/137 Test: test_TimeConversions

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_TimeConversions.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/BasicAstrodynamics

"test_TimeConversions" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 5 test cases...

*** No errors detected

<end of output>

Test time = 1.54 sec

Test Passed.

"test_TimeConversions" end time: Feb 17 12:25 W. Europe Standard Time

"test_TimeConversions" time elapsed: 00:00:01

22/137 Testing: test_CelestialBodyConstants

22/137 Test: test_CelestialBodyConstants

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_CelestialBodyConstants.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/BasicAstrodynamics

"test_CelestialBodyConstants" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 4 test cases...

*** No errors detected

<end of output>

Test time = 1.84 sec

Test Passed.

"test_CelestialBodyConstants" end time: Feb 17 12:25 W. Europe Standard Time

"test_CelestialBodyConstants" time elapsed: 00:00:01

23/137 Testing: test_ModifiedEquinoctialElementConversions

23/137 Test: test_ModifiedEquinoctialElementConversions

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_ModifiedEquinoctialElementConversions.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/BasicAstrodynamics

"test_ModifiedEquinoctialElementConversions" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 4 test cases...

*** No errors detected

<end of output>

Test time = 1.77 sec

Test Passed.

"test_ModifiedEquinoctialElementConversions" end time: Feb 17 12:25 W. Europe Standard Time

"test_ModifiedEquinoctialElementConversions" time elapsed: 00:00:01

24/137 Testing: test_GeodeticCoordinateConversions

24/137 Test: test_GeodeticCoordinateConversions

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_GeodeticCoordinateConversions.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/BasicAstrodynamics

"test_GeodeticCoordinateConversions" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 1.76 sec

Test Passed.

"test_GeodeticCoordinateConversions" end time: Feb 17 12:25 W. Europe Standard Time

"test_GeodeticCoordinateConversions" time elapsed: 00:00:01

25/137 Testing: test_StateElementConversions

25/137 Test: test_StateElementConversions

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_StateElementConversions.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/BasicAstrodynamics

"test_StateElementConversions" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 1.65 sec

Test Passed.

"test_StateElementConversions" end time: Feb 17 12:25 W. Europe Standard Time

"test_StateElementConversions" time elapsed: 00:00:01

26/137 Testing: test_SphericalOrbitalStateConversions

26/137 Test: test_SphericalOrbitalStateConversions

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_SphericalOrbitalStateConversions.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/BasicAstrodynamics

"test_SphericalOrbitalStateConversions" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 1.80 sec

Test Passed.

"test_SphericalOrbitalStateConversions" end time: Feb 17 12:25 W. Europe Standard Time

"test_SphericalOrbitalStateConversions" time elapsed: 00:00:01

27/137 Testing: test_BodyShapeModels

27/137 Test: test_BodyShapeModels

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_BodyShapeModels.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/BasicAstrodynamics

"test_BodyShapeModels" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 1.90 sec

Test Passed.

"test_BodyShapeModels" end time: Feb 17 12:25 W. Europe Standard Time

"test_BodyShapeModels" time elapsed: 00:00:01

28/137 Testing: test_UnifiedStateModelElementConversions

28/137 Test: test_UnifiedStateModelElementConversions

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_UnifiedStateModelElementConversions.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/BasicAstrodynamics

"test_UnifiedStateModelElementConversions" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 2 test cases...

*** No errors detected

<end of output>

Test time = 4.33 sec

Test Passed.

"test_UnifiedStateModelElementConversions" end time: Feb 17 12:25 W. Europe Standard Time

"test_UnifiedStateModelElementConversions" time elapsed: 00:00:04

29/137 Testing: test_CannonBallRadiationPressureAccelerationAndForce

29/137 Test: test_CannonBallRadiationPressureAccelerationAndForce

Command:

"C:/tudatBundle/tudat/bin/unit_tests/test_CannonBallRadiationPressureAccelerationAndForce.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/ElectroMagnetism

"test_CannonBallRadiationPressureAccelerationAndForce" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 9 test cases...

*** No errors detected

<end of output>

Test time = 1.80 sec

Test Passed.

"test_CannonBallRadiationPressureAccelerationAndForce" end time: Feb 17 12:25 W. Europe Standard Time

"test_CannonBallRadiationPressureAccelerationAndForce" time elapsed: 00:00:01

30/137 Testing: test_LorentzStaticMagneticAccelerationAndForce

30/137 Test: test_LorentzStaticMagneticAccelerationAndForce

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_LorentzStaticMagneticAccelerationAndForce.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/ElectroMagnetism

"test_LorentzStaticMagneticAccelerationAndForce" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 5 test cases...

*** No errors detected

<end of output>

Test time = 1.53 sec

Test Passed.

"test_LorentzStaticMagneticAccelerationAndForce" end time: Feb 17 12:25 W. Europe Standard Time

"test_LorentzStaticMagneticAccelerationAndForce" time elapsed: 00:00:01

31/137 Testing: test_RadiationPressureInterface

31/137 Test: test_RadiationPressureInterface

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_RadiationPressureInterface.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/ElectroMagnetism

"test_RadiationPressureInterface" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 2 test cases...

*** No errors detected

<end of output>

Test time = 1.51 sec

Test Passed.

"test_RadiationPressureInterface" end time: Feb 17 12:25 W. Europe Standard Time

"test_RadiationPressureInterface" time elapsed: 00:00:01

32/137 Testing: test_ApproximatePlanetPositions

32/137 Test: test_ApproximatePlanetPositions

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_ApproximatePlanetPositions.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Ephemerides

"test_ApproximatePlanetPositions" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 2 test cases...

*** No errors detected

<end of output>

Test time = 1.47 sec

Test Passed.

"test_ApproximatePlanetPositions" end time: Feb 17 12:25 W. Europe Standard Time

"test_ApproximatePlanetPositions" time elapsed: 00:00:01

33/137 Testing: test_TabulatedEphemeris

33/137 Test: test_TabulatedEphemeris

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_TabulatedEphemeris.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Ephemerides

"test_TabulatedEphemeris" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 2.41 sec

Test Passed.

"test_TabulatedEphemeris" end time: Feb 17 12:25 W. Europe Standard Time

"test_TabulatedEphemeris" time elapsed: 00:00:02

34/137 Testing: test_CartesianStateExtractor

34/137 Test: test_CartesianStateExtractor

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_CartesianStateExtractor.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Ephemerides

"test_CartesianStateExtractor" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 7 test cases...

*** No errors detected

<end of output>

Test time = 1.55 sec

Test Passed.

"test_CartesianStateExtractor" end time: Feb 17 12:25 W. Europe Standard Time

"test_CartesianStateExtractor" time elapsed: 00:00:01

35/137 Testing: test_KeplerStateExtractor

35/137 Test: test_KeplerStateExtractor

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_KeplerStateExtractor.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Ephemerides

"test_KeplerStateExtractor" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 8 test cases...

*** No errors detected

<end of output>

Test time = 1.59 sec

Test Passed.

"test_KeplerStateExtractor" end time: Feb 17 12:25 W. Europe Standard Time

"test_KeplerStateExtractor" time elapsed: 00:00:01

36/137 Testing: test_RotationalEphemeris

36/137 Test: test_RotationalEphemeris

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_RotationalEphemeris.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Ephemerides

"test_RotationalEphemeris" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 4.49 sec

Test Passed.

"test_RotationalEphemeris" end time: Feb 17 12:25 W. Europe Standard Time

"test_RotationalEphemeris" time elapsed: 00:00:04

37/137 Testing: test_SimpleRotationalEphemeris

37/137 Test: test_SimpleRotationalEphemeris

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_SimpleRotationalEphemeris.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Ephemerides

"test_SimpleRotationalEphemeris" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 1.71 sec

Test Passed.

"test_SimpleRotationalEphemeris" end time: Feb 17 12:25 W. Europe Standard Time

"test_SimpleRotationalEphemeris" time elapsed: 00:00:01

38/137 Testing: test_FrameManager

38/137 Test: test_FrameManager

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_FrameManager.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Ephemerides

"test_FrameManager" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 1.99 sec

Test Passed.

"test_FrameManager" end time: Feb 17 12:25 W. Europe Standard Time

"test_FrameManager" time elapsed: 00:00:01

39/137 Testing: test_CompositeEphemeris

39/137 Test: test_CompositeEphemeris

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_CompositeEphemeris.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Ephemerides

"test_CompositeEphemeris" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 2.54 sec

Test Passed.

"test_CompositeEphemeris" end time: Feb 17 12:25 W. Europe Standard Time

"test_CompositeEphemeris" time elapsed: 00:00:02

40/137 Testing: test_KeplerEphemeris

40/137 Test: test_KeplerEphemeris

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_KeplerEphemeris.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Ephemerides

"test_KeplerEphemeris" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 2 test cases...

*** No errors detected

<end of output>

Test time = 1.64 sec

Test Passed.

"test_KeplerEphemeris" end time: Feb 17 12:25 W. Europe Standard Time

"test_KeplerEphemeris" time elapsed: 00:00:01

41/137 Testing: test_SphericalHarmonicsGravityField

41/137 Test: test_SphericalHarmonicsGravityField

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_SphericalHarmonicsGravityField.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Gravitation

"test_SphericalHarmonicsGravityField" start time: Feb 17 12:25 W. Europe Standard Time

Output:

Running 4 test cases...

*** No errors detected

<end of output>

Test time = 1.65 sec

Test Passed.

"test_SphericalHarmonicsGravityField" end time: Feb 17 12:26 W. Europe Standard Time

"test_SphericalHarmonicsGravityField" time elapsed: 00:00:01

42/137 Testing: test_GravitationalForce

42/137 Test: test_GravitationalForce

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_GravitationalForce.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Gravitation

"test_GravitationalForce" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 1.99 sec

Test Passed.

"test_GravitationalForce" end time: Feb 17 12:26 W. Europe Standard Time

"test_GravitationalForce" time elapsed: 00:00:01

43/137 Testing: test_CentralAndZonalGravityModel

43/137 Test: test_CentralAndZonalGravityModel

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_CentralAndZonalGravityModel.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Gravitation

"test_CentralAndZonalGravityModel" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 5 test cases...

*** No errors detected

<end of output>

Test time = 1.93 sec

Test Passed.

"test_CentralAndZonalGravityModel" end time: Feb 17 12:26 W. Europe Standard Time

"test_CentralAndZonalGravityModel" time elapsed: 00:00:01

44/137 Testing: test_UnitConversionsCircularRestrictedThreeBodyProblem

44/137 Test: test_UnitConversionsCircularRestrictedThreeBodyProblem

Command:

"C:/tudatBundle/tudat/bin/unit_tests/test_UnitConversionsCircularRestrictedThreeBodyProblem.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Gravitation

"test_UnitConversionsCircularRestrictedThreeBodyProblem" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 2 test cases...

*** No errors detected

<end of output>

Test time = 1.49 sec

Test Passed.

"test_UnitConversionsCircularRestrictedThreeBodyProblem" end time: Feb 17 12:26 W. Europe Standard Time

"test_UnitConversionsCircularRestrictedThreeBodyProblem" time elapsed: 00:00:01

45/137 Testing: test_JacobiEnergy

45/137 Test: test_JacobiEnergy

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_JacobiEnergy.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Gravitation

"test_JacobiEnergy" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 1.75 sec

Test Passed.

"test_JacobiEnergy" end time: Feb 17 12:26 W. Europe Standard Time

"test_JacobiEnergy" time elapsed: 00:00:01

46/137 Testing: test_LibrationPoints

46/137 Test: test_LibrationPoints

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_LibrationPoints.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Gravitation

"test_LibrationPoints" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 6 test cases...

*** No errors detected

<end of output>

Test time = 1.99 sec

Test Passed.

"test_LibrationPoints" end time: Feb 17 12:26 W. Europe Standard Time

"test_LibrationPoints" time elapsed: 00:00:01

47/137 Testing: test_StateDerivativeRestrictedThreeBodyProblem

47/137 Test: test_StateDerivativeRestrictedThreeBodyProblem

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_StateDerivativeRestrictedThreeBodyProblem.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Gravitation

"test_StateDerivativeRestrictedThreeBodyProblem" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 1.90 sec

Test Passed.

"test_StateDerivativeRestrictedThreeBodyProblem" end time: Feb 17 12:26 W. Europe Standard Time

"test_StateDerivativeRestrictedThreeBodyProblem" time elapsed: 00:00:01

48/137 Testing: test_SphericalHarmonicsGravityModel

48/137 Test: test_SphericalHarmonicsGravityModel

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_SphericalHarmonicsGravityModel.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Gravitation

"test_SphericalHarmonicsGravityModel" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 5 test cases...

*** No errors detected

<end of output>

Test time = 2.15 sec

Test Passed.

"test_SphericalHarmonicsGravityModel" end time: Feb 17 12:26 W. Europe Standard Time

"test_SphericalHarmonicsGravityModel" time elapsed: 00:00:02

49/137 Testing: test_ThirdBodyPerturbation

49/137 Test: test_ThirdBodyPerturbation

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_ThirdBodyPerturbation.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Gravitation

"test_ThirdBodyPerturbation" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 2 test cases...

*** No errors detected

<end of output>

Test time = 1.80 sec

Test Passed.

"test_ThirdBodyPerturbation" end time: Feb 17 12:26 W. Europe Standard Time

"test_ThirdBodyPerturbation" time elapsed: 00:00:01

50/137 Testing: test_TriAxialEllipsoidGravity

50/137 Test: test_TriAxialEllipsoidGravity

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_TriAxialEllipsoidGravity.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Gravitation

"test_TriAxialEllipsoidGravity" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 1.80 sec

Test Passed.

"test_TriAxialEllipsoidGravity" end time: Feb 17 12:26 W. Europe Standard Time

"test_TriAxialEllipsoidGravity" time elapsed: 00:00:01

51/137 Testing: test_GravityFieldVariations

51/137 Test: test_GravityFieldVariations

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_GravityFieldVariations.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Gravitation

"test_GravityFieldVariations" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 4.98 sec

Test Passed.

"test_GravityFieldVariations" end time: Feb 17 12:26 W. Europe Standard Time

"test_GravityFieldVariations" time elapsed: 00:00:04

52/137 Testing: test_MutualSphericalHarmonicsGravityModel

52/137 Test: test_MutualSphericalHarmonicsGravityModel

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_MutualSphericalHarmonicsGravityModel.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Gravitation

"test_MutualSphericalHarmonicsGravityModel" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 3.16 sec

Test Passed.

"test_MutualSphericalHarmonicsGravityModel" end time: Feb 17 12:26 W. Europe Standard Time

"test_MutualSphericalHarmonicsGravityModel" time elapsed: 00:00:03

53/137 Testing: test_EscapeAndCapture

53/137 Test: test_EscapeAndCapture

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_EscapeAndCapture.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/MissionSegments

"test_EscapeAndCapture" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 3 test cases...

*** No errors detected

<end of output>

Test time = 1.70 sec

Test Passed.

"test_EscapeAndCapture" end time: Feb 17 12:26 W. Europe Standard Time

"test_EscapeAndCapture" time elapsed: 00:00:01

54/137 Testing: test_GravityAssist

54/137 Test: test_GravityAssist

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_GravityAssist.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/MissionSegments

"test_GravityAssist" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 14 test cases...

*** No errors detected

<end of output>

Test time = 2.04 sec

Test Passed.

"test_GravityAssist" end time: Feb 17 12:26 W. Europe Standard Time

"test_GravityAssist" time elapsed: 00:00:02

55/137 Testing: test_LambertTargeterIzzo

55/137 Test: test_LambertTargeterIzzo

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_LambertTargeterIzzo.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/MissionSegments

"test_LambertTargeterIzzo" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 4 test cases...

*** No errors detected

<end of output>

Test time = 2.09 sec

Test Passed.

"test_LambertTargeterIzzo" end time: Feb 17 12:26 W. Europe Standard Time

"test_LambertTargeterIzzo" time elapsed: 00:00:02

56/137 Testing: test_LambertTargeterGooding

56/137 Test: test_LambertTargeterGooding

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_LambertTargeterGooding.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/MissionSegments

"test_LambertTargeterGooding" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 2 test cases...

*** No errors detected

<end of output>

Test time = 1.94 sec

Test Passed.

"test_LambertTargeterGooding" end time: Feb 17 12:26 W. Europe Standard Time

"test_LambertTargeterGooding" time elapsed: 00:00:01

57/137 Testing: test_LambertTargeter

57/137 Test: test_LambertTargeter

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_LambertTargeter.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/MissionSegments

"test_LambertTargeter" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 2.06 sec

Test Passed.

"test_LambertTargeter" end time: Feb 17 12:26 W. Europe Standard Time

"test_LambertTargeter" time elapsed: 00:00:02

58/137 Testing: test_LambertRoutines

58/137 Test: test_LambertRoutines

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_LambertRoutines.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/MissionSegments

"test_LambertRoutines" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 13 test cases...

*** No errors detected

<end of output>

Test time = 1.94 sec

Test Passed.

"test_LambertRoutines" end time: Feb 17 12:26 W. Europe Standard Time

"test_LambertRoutines" time elapsed: 00:00:01

59/137 Testing: test_ZeroRevolutionLambertTargeterIzzo

59/137 Test: test_ZeroRevolutionLambertTargeterIzzo

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_ZeroRevolutionLambertTargeterIzzo.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/MissionSegments

"test_ZeroRevolutionLambertTargeterIzzo" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 4 test cases...

*** No errors detected

<end of output>

Test time = 1.76 sec

Test Passed.

"test_ZeroRevolutionLambertTargeterIzzo" end time: Feb 17 12:26 W. Europe Standard Time

"test_ZeroRevolutionLambertTargeterIzzo" time elapsed: 00:00:01

60/137 Testing: test_MultiRevolutionLambertTargeterIzzo

60/137 Test: test_MultiRevolutionLambertTargeterIzzo

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_MultiRevolutionLambertTargeterIzzo.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/MissionSegments

"test_MultiRevolutionLambertTargeterIzzo" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 3 test cases...

*** No errors detected

<end of output>

Test time = 2.32 sec

Test Passed.

"test_MultiRevolutionLambertTargeterIzzo" end time: Feb 17 12:26 W. Europe Standard Time

"test_MultiRevolutionLambertTargeterIzzo" time elapsed: 00:00:02

61/137 Testing: test_MathematicalShapeFunctions

61/137 Test: test_MathematicalShapeFunctions

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_MathematicalShapeFunctions.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/MissionSegments

"test_MathematicalShapeFunctions" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 3 test cases...

*** No errors detected

<end of output>

Test time = 2.05 sec

Test Passed.

"test_MathematicalShapeFunctions" end time: Feb 17 12:26 W. Europe Standard Time

"test_MathematicalShapeFunctions" time elapsed: 00:00:02

62/137 Testing: test_ReferenceFrameTransformations

62/137 Test: test_ReferenceFrameTransformations

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_ReferenceFrameTransformations.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/ReferenceFrames

"test_ReferenceFrameTransformations" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 10 test cases...

*** No errors detected

<end of output>

Test time = 2.12 sec

Test Passed.

"test_ReferenceFrameTransformations" end time: Feb 17 12:26 W. Europe Standard Time

"test_ReferenceFrameTransformations" time elapsed: 00:00:02

63/137 Testing: test_ApparentAccelerationModel

63/137 Test: test_ApparentAccelerationModel

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_ApparentAccelerationModel.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/ReferenceFrames

"test_ApparentAccelerationModel" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 5 test cases...

*** No errors detected

<end of output>

Test time = 1.81 sec

Test Passed.

"test_ApparentAccelerationModel" end time: Feb 17 12:26 W. Europe Standard Time

"test_ApparentAccelerationModel" time elapsed: 00:00:01

64/137 Testing: test_AerodynamicAngleCalculator

64/137 Test: test_AerodynamicAngleCalculator

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_AerodynamicAngleCalculator.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/ReferenceFrames

"test_AerodynamicAngleCalculator" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 1.66 sec

Test Passed.

"test_AerodynamicAngleCalculator" end time: Feb 17 12:26 W. Europe Standard Time

"test_AerodynamicAngleCalculator" time elapsed: 00:00:01

65/137 Testing: test_LightTime

65/137 Test: test_LightTime

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_LightTime.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/ObservationModels

"test_LightTime" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 2.30 sec

Test Passed.

"test_LightTime" end time: Feb 17 12:26 W. Europe Standard Time

"test_LightTime" time elapsed: 00:00:02

66/137 Testing: test_AngularPositionModel

66/137 Test: test_AngularPositionModel

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_AngularPositionModel.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/ObservationModels

"test_AngularPositionModel" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 1 test case...

Warning, position of Mars taken as barycenter of that body's planetary system.

*** No errors detected

<end of output>

Test time = 2.60 sec

Test Passed.

"test_AngularPositionModel" end time: Feb 17 12:26 W. Europe Standard Time

"test_AngularPositionModel" time elapsed: 00:00:02

67/137 Testing: test_OneWayRangeModel

67/137 Test: test_OneWayRangeModel

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_OneWayRangeModel.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/ObservationModels

"test_OneWayRangeModel" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 1 test case...

Warning, position of Mars taken as barycenter of that body's planetary system.

*** No errors detected

<end of output>

Test time = 3.43 sec

Test Passed.

"test_OneWayRangeModel" end time: Feb 17 12:26 W. Europe Standard Time

"test_OneWayRangeModel" time elapsed: 00:00:03

68/137 Testing: test_PositionObservationModel

68/137 Test: test_PositionObservationModel

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_PositionObservationModel.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/ObservationModels

"test_PositionObservationModel" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 2.50 sec

Test Passed.

"test_PositionObservationModel" end time: Feb 17 12:26 W. Europe Standard Time

"test_PositionObservationModel" time elapsed: 00:00:02

69/137 Testing: test_AccelerationPartials

69/137 Test: test_AccelerationPartials

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_AccelerationPartials.exe"

Directory: C:/build-tudatBundle-Desktop-
Default/tudat/Tudat/Astrodynamics/OrbitDetermination/AccelerationPartials

"test_AccelerationPartials" start time: Feb 17 12:26 W. Europe Standard Time

Output:

Running 3 test cases...

*** No errors detected

<end of output>

Test time = 2.91 sec

Test Passed.

"test_AccelerationPartials" end time: Feb 17 12:27 W. Europe Standard Time

"test_AccelerationPartials" time elapsed: 00:00:02

70/137 Testing: test_SphericalHarmonicPartials

70/137 Test: test_SphericalHarmonicPartials

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_SphericalHarmonicPartials.exe"

Directory: C:/build-tudatBundle-Desktop-
Default/tudat/Tudat/Astrodynamics/OrbitDetermination/AccelerationPartials

"test_SphericalHarmonicPartials" start time: Feb 17 12:27 W. Europe Standard Time

Output:

Running 2 test cases...

*** No errors detected

<end of output>

Test time = 2.78 sec

Test Passed.

"test_SphericalHarmonicPartials" end time: Feb 17 12:27 W. Europe Standard Time

"test_SphericalHarmonicPartials" time elapsed: 00:00:02

71/137 Testing: test_RotationMatrixPartials

71/137 Test: test_RotationMatrixPartials

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_RotationMatrixPartials.exe"

Directory: C:/build-tudatBundle-Desktop-
Default/tudat/Tudat/Astrodynamics/OrbitDetermination/ObservationPartials

"test_RotationMatrixPartials" start time: Feb 17 12:27 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 2.34 sec

Test Passed.

"test_RotationMatrixPartials" end time: Feb 17 12:27 W. Europe Standard Time

"test_RotationMatrixPartials" time elapsed: 00:00:02

72/137 Testing: test_CentralBodyData

72/137 Test: test_CentralBodyData

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_CentralBodyData.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Propagators

"test_CentralBodyData" start time: Feb 17 12:27 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 2.08 sec

Test Passed.

"test_CentralBodyData" end time: Feb 17 12:27 W. Europe Standard Time

"test_CentralBodyData" time elapsed: 00:00:02

73/137 Testing: test_CowellStateDerivative

73/137 Test: test_CowellStateDerivative

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_CowellStateDerivative.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Propagators

"test_CowellStateDerivative" start time: Feb 17 12:27 W. Europe Standard Time

Output:

Running 2 test cases...

Warning, position of Mars taken as barycenter of that body's planetary system.

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

*** No errors detected

<end of output>

Test time = 22.11 sec

Test Passed.

"test_CowellStateDerivative" end time: Feb 17 12:27 W. Europe Standard Time

"test_CowellStateDerivative" time elapsed: 00:00:22

74/137 Testing: test_EnckeStateDerivative

74/137 Test: test_EnckeStateDerivative

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_EnckeStateDerivative.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Propagators

"test_EnckeStateDerivative" start time: Feb 17 12:27 W. Europe Standard Time

Output:

Running 3 test cases...

Warning, position of Jupiter taken as barycenter of that body's planetary system.

Warning, position of Mars taken as barycenter of that body's planetary system.

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, position of Jupiter taken as barycenter of that body's planetary system.

Warning, position of Mars taken as barycenter of that body's planetary system.

Warning, tabulated ephemeris is being reset using data at different precision
Warning, tabulated ephemeris is being reset using data at different precision
Warning, tabulated ephemeris is being reset using data at different precision
Warning, tabulated ephemeris is being reset using data at different precision
Warning, tabulated ephemeris is being reset using data at different precision
Warning, tabulated ephemeris is being reset using data at different precision
Warning, position of Mars taken as barycenter of that body's planetary system.
Warning, tabulated ephemeris is being reset using data at different precision
Warning, tabulated ephemeris is being reset using data at different precision
Warning, position of Mars taken as barycenter of that body's planetary system.
Warning, tabulated ephemeris is being reset using data at different precision
Warning, tabulated ephemeris is being reset using data at different precision
Warning, position of Mars taken as barycenter of that body's planetary system.
Warning, tabulated ephemeris is being reset using data at different precision
Warning, tabulated ephemeris is being reset using data at different precision
Warning, position of Mars taken as barycenter of that body's planetary system.
Warning, tabulated ephemeris is being reset using data at different precision
Warning, tabulated ephemeris is being reset using data at different precision
Testint with eccentricity 0.8Testint with eccentricity 0.9Testint with eccentricity 0.95Testint with eccentricity 0.99

*** No errors detected

<end of output>

Test time = 32.71 sec

Test Passed.

"test_EnckeStateDerivative" end time: Feb 17 12:28 W. Europe Standard Time

"test_EnckeStateDerivative" time elapsed: 00:00:32

75/137 Testing: test_SequentialVariationEquationIntegration

75/137 Test: test_SequentialVariationEquationIntegration

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_SequentialVariationEquationIntegration.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Propagators

"test_SequentialVariationEquationIntegration" start time: Feb 17 12:28 W. Europe Standard Time

Output:

Running 1 test case...

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

*** No errors detected

<end of output>

Test time = 16.24 sec

Test Passed.

"test_SequentialVariationEquationIntegration" end time: Feb 17 12:28 W. Europe Standard Time

"test_SequentialVariationEquationIntegration" time elapsed: 00:00:16

76/137 Testing: test_VariationalEquations

76/137 Test: test_VariationalEquations

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_VariationalEquations.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Propagators

"test_VariationalEquations" start time: Feb 17 12:28 W. Europe Standard Time

Output:

Running 1 test case...

Warning, position of Mars taken as barycenter of that body's planetary system.

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, position of Mars taken as barycenter of that body's planetary system.

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, position of Mars taken as barycenter of that body's planetary system.

Warning, tabulated ephemeris is being reset using data at different precision

Warning, position of Mars taken as barycenter of that body's planetary system.

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, position of Mars taken as barycenter of that body's planetary system.

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, position of Mars taken as barycenter of that body's planetary system.

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, position of Mars taken as barycenter of that body's planetary system.

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, position of Mars taken as barycenter of that body's planetary system.

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, position of Mars taken as barycenter of that body's planetary system.

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, position of Mars taken as barycenter of that body's planetary system.

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, position of Mars taken as barycenter of that body's planetary system.

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, position of Mars taken as barycenter of that body's planetary system.

Warning, tabulated ephemeris is being reset using data at different precision
Warning, position of Mars taken as barycenter of that body's planetary system.
Warning, tabulated ephemeris is being reset using data at different precision
Warning, tabulated ephemeris is being reset using data at different precision
Warning, position of Mars taken as barycenter of that body's planetary system.
Warning, tabulated ephemeris is being reset using data at different precision
Warning, tabulated ephemeris is being reset using data at different precision
Warning, position of Mars taken as barycenter of that body's planetary system.
Warning, tabulated ephemeris is being reset using data at different precision
Warning, tabulated ephemeris is being reset using data at different precision
Warning, position of Mars taken as barycenter of that body's planetary system.
Warning, tabulated ephemeris is being reset using data at different precision
Warning, tabulated ephemeris is being reset using data at different precision
Warning, position of Mars taken as barycenter of that body's planetary system.
Warning, tabulated ephemeris is being reset using data at different precision
Warning, tabulated ephemeris is being reset using data at different precision

*** No errors detected

<end of output>

Test time = 42.98 sec

Test Passed.

"test_VariationalEquations" end time: Feb 17 12:29 W. Europe Standard Time

"test_VariationalEquations" time elapsed: 00:00:42

77/137 Testing: test_EnvironmentModelUpdater

77/137 Test: test_EnvironmentModelUpdater

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_EnvironmentModelUpdater.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Propagators

"test_EnvironmentModelUpdater" start time: Feb 17 12:29 W. Europe Standard Time

Output:

<end of output>

Test time = 0.00 sec

Test Failed.

"test_EnvironmentModelUpdater" end time: Feb 17 12:29 W. Europe Standard Time

"test_EnvironmentModelUpdater" time elapsed: 00:00:00

78/137 Testing: test_BodyMassPropagation

78/137 Test: test_BodyMassPropagation

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_BodyMassPropagation.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Propagators

"test_BodyMassPropagation" start time: Feb 17 12:29 W. Europe Standard Time

Output:

Running 2 test cases...

*** No errors detected

<end of output>

Test time = 3.05 sec

Test Passed.

"test_BodyMassPropagation" end time: Feb 17 12:29 W. Europe Standard Time

"test_BodyMassPropagation" time elapsed: 00:00:03

79/137 Testing: test_MultiTypeStatePropagation

79/137 Test: test_MultiTypeStatePropagation

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_MultiTypeStatePropagation.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Propagators

"test_MultiTypeStatePropagation" start time: Feb 17 12:29 W. Europe Standard Time

Output:

Running 1 test case...

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

Warning, tabulated ephemeris is being reset using data at different precision

*** No errors detected

<end of output>

Test time = 3.81 sec

Test Passed.

"test_MultiTypeStatePropagation" end time: Feb 17 12:29 W. Europe Standard Time

"test_MultiTypeStatePropagation" time elapsed: 00:00:03

80/137 Testing: test_DependentVariableOutput

80/137 Test: test_DependentVariableOutput

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_DependentVariableOutput.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Propagators

"test_DependentVariableOutput" start time: Feb 17 12:29 W. Europe Standard Time

Output:

Running 1 test case...

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Mach number of Apollo

1, Altitude of Apollo w.r.t. Earth

2, Relative distance of Apollo w.r.t. Earth

3, Relative speed of Apollo w.r.t. Earth

4, Single acceleration norm of type central gravity , acting on Apollo, exerted by Earth

- 5, Aerodynamic g-load of Apollo w.r.t. Earth
- 6, Stagnation-point heat flux of Apollo w.r.t. Earth
- 7, Local freestream temperature of Apollo w.r.t. Earth
- 8, Geodetic latitude of Apollo w.r.t. Earth
- 9, Density of Apollo w.r.t. Earth
- 10, Body orientation angle latitude angle of Apollo
- 11, Body orientation angle longitude angle of Apollo
- 12, Body orientation angle angle of attack of Apollo
- 13, Body orientation angle sideslip angle of Apollo
- 14, Body orientation angle bank angle of Apollo
- 15, Relative position of Apollo w.r.t. Earth
- 18, Relative velocity of Apollo w.r.t. Earth
- 21, Single acceleration in inertial frame of type central gravity , acting on Apollo, exerted by Earth
- 24, Total acceleration in inertial frame of Apollo
- 27, Aerodynamic moment coefficients of Apollo
- 30, Aerodynamic force coefficients of Apollo
- 33, Single acceleration in inertial frame of type aerodynamic , acting on Apollo, exerted by Earth

Propagated

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

- 0, Mach number of Apollo
- 1, Altitude of Apollo w.r.t. Earth
- 2, Relative distance of Apollo w.r.t. Earth
- 3, Relative speed of Apollo w.r.t. Earth
- 4, Single acceleration norm of type central gravity , acting on Apollo, exerted by Earth
- 5, Aerodynamic g-load of Apollo w.r.t. Earth
- 6, Stagnation-point heat flux of Apollo w.r.t. Earth
- 7, Local freestream temperature of Apollo w.r.t. Earth
- 8, Geodetic latitude of Apollo w.r.t. Earth
- 9, Density of Apollo w.r.t. Earth
- 10, Body orientation angle latitude angle of Apollo
- 11, Body orientation angle longitude angle of Apollo

- 12, Body orientation angle angle of attack of Apollo
- 13, Body orientation angle sideslip angle of Apollo
- 14, Body orientation angle bank angle of Apollo
- 15, Relative position of Apollo w.r.t. Earth
- 18, Relative velocity of Apollo w.r.t. Earth
- 21, Single acceleration in inertial frame of type central gravity , acting on Apollo, exerted by Earth
- 24, Total acceleration in inertial frame of Apollo
- 27, Aerodynamic moment coefficients of Apollo
- 30, Aerodynamic force coefficients of Apollo
- 33, Single acceleration in inertial frame of type aerodynamic , acting on Apollo, exerted by Earth

Propagated

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

- 0, Mach number of Apollo
- 1, Altitude of Apollo w.r.t. Earth
- 2, Relative distance of Apollo w.r.t. Earth
- 3, Relative speed of Apollo w.r.t. Earth
- 4, Single acceleration norm of type central gravity , acting on Apollo, exerted by Earth
- 5, Aerodynamic g-load of Apollo w.r.t. Earth
- 6, Stagnation-point heat flux of Apollo w.r.t. Earth
- 7, Local freestream temperature of Apollo w.r.t. Earth
- 8, Geodetic latitude of Apollo w.r.t. Earth
- 9, Density of Apollo w.r.t. Earth
- 10, Body orientation angle latitude angle of Apollo
- 11, Body orientation angle longitude angle of Apollo
- 12, Body orientation angle angle of attack of Apollo
- 13, Body orientation angle sideslip angle of Apollo
- 14, Body orientation angle bank angle of Apollo
- 15, Relative position of Apollo w.r.t. Earth
- 18, Relative velocity of Apollo w.r.t. Earth
- 21, Single acceleration in inertial frame of type central gravity , acting on Apollo, exerted by Earth
- 24, Total acceleration in inertial frame of Apollo

27, Aerodynamic moment coefficients of Apollo

30, Aerodynamic force coefficients of Apollo

33, Single acceleration in inertial frame of type aerodynamic , acting on Apollo, exerted by Earth

Propagated

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Mach number of Apollo

1, Altitude of Apollo w.r.t. Earth

2, Relative distance of Apollo w.r.t. Earth

3, Relative speed of Apollo w.r.t. Earth

4, Single acceleration norm of type central gravity , acting on Apollo, exerted by Earth

5, Aerodynamic g-load of Apollo w.r.t. Earth

6, Stagnation-point heat flux of Apollo w.r.t. Earth

7, Local freestream temperature of Apollo w.r.t. Earth

8, Geodetic latitude of Apollo w.r.t. Earth

9, Density of Apollo w.r.t. Earth

10, Body orientation angle latitude angle of Apollo

11, Body orientation angle longitude angle of Apollo

12, Body orientation angle angle of attack of Apollo

13, Body orientation angle sideslip angle of Apollo

14, Body orientation angle bank angle of Apollo

15, Relative position of Apollo w.r.t. Earth

18, Relative velocity of Apollo w.r.t. Earth

21, Single acceleration in inertial frame of type central gravity , acting on Apollo, exerted by Earth

24, Total acceleration in inertial frame of Apollo

27, Aerodynamic moment coefficients of Apollo

30, Aerodynamic force coefficients of Apollo

33, Single acceleration in inertial frame of type aerodynamic , acting on Apollo, exerted by Earth

Propagated

*** No errors detected

<end of output>

Test time = 18.94 sec

Test Passed.

"test_DependentVariableOutput" end time: Feb 17 12:29 W. Europe Standard Time

"test_DependentVariableOutput" time elapsed: 00:00:18

81/137 Testing: test_StoppingConditions

81/137 Test: test_StoppingConditions

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_StoppingConditions.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Propagators

"test_StoppingConditions" start time: Feb 17 12:29 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 7.16 sec

Test Passed.

"test_StoppingConditions" end time: Feb 17 12:29 W. Europe Standard Time

"test_StoppingConditions" time elapsed: 00:00:07

82/137 Testing: test_CustomStatePropagation

82/137 Test: test_CustomStatePropagation

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_CustomStatePropagation.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Propagators

"test_CustomStatePropagation" start time: Feb 17 12:29 W. Europe Standard Time

Output:

Running 5 test cases...

Warning, tabulated ephemeris is being reset using data at different precision

*** No errors detected

<end of output>

Test time = 2.84 sec

Test Passed.

"test_CustomStatePropagation" end time: Feb 17 12:29 W. Europe Standard Time

"test_CustomStatePropagation" time elapsed: 00:00:02

83/137 Testing: test_ShapiroTimeDelay

83/137 Test: test_ShapiroTimeDelay

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_ShapiroTimeDelay.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Relativity

"test_ShapiroTimeDelay" start time: Feb 17 12:29 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 1.94 sec

Test Passed.

"test_ShapiroTimeDelay" end time: Feb 17 12:29 W. Europe Standard Time

"test_ShapiroTimeDelay" time elapsed: 00:00:01

84/137 Testing: test_ThrustAcceleration

84/137 Test: test_ThrustAcceleration

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_ThrustAcceleration.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/Propulsion

"test_ThrustAcceleration" start time: Feb 17 12:29 W. Europe Standard Time

Output:

Running 8 test cases...

Warning, cannot reset dependentOrientationCalculator, incompatible object already exists

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration in inertial frame of type thrust , acting on Vehicle

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Mach number of Apollo

1, Body orientation angle angle of attack of Apollo

2, Body orientation angle sideslip angle of Apollo

3, Body orientation angle bank angle of Apollo

4, Airspeed of Apollo w.r.t. Earth

5, Density of Apollo w.r.t. Earth

6, Rotation matrix from inertial frame to body frame of Apollo

15, Rotation matrix to body-fixed frame of Apollo

24, Single acceleration in inertial frame of type aerodynamic , acting on Apollo, exerted by Earth

27, Single acceleration in inertial frame of type thrust , acting on Apollo

30, Aerodynamic force coefficients of Apollo

33, Aerodynamic moment coefficients of Apollo

36, Rotation matrix from inertial frame to aerodynamic frame of Apollo

45, Rotation matrix from aerodynamic frame to body frame of Apollo

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Mach number of Apollo

- 1, Body orientation angle angle of attack of Apollo
- 2, Body orientation angle sideslip angle of Apollo
- 3, Body orientation angle bank angle of Apollo
- 4, Airspeed of Apollo w.r.t. Earth
- 5, Density of Apollo w.r.t. Earth
- 6, Rotation matrix from inertial frame to body frame of Apollo
- 15, Rotation matrix to body-fixed frame of Apollo
- 24, Single acceleration in inertial frame of type aerodynamic , acting on Apollo, exerted by Earth
- 27, Single acceleration in inertial frame of type thrust , acting on Apollo
- 30, Aerodynamic force coefficients of Apollo
- 33, Aerodynamic moment coefficients of Apollo
- 36, Rotation matrix from inertial frame to aerodynamic frame of Apollo
- 45, Rotation matrix from aerodynamic frame to body frame of Apollo

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

- 0, Single acceleration in inertial frame of type thrust , acting on Asterix
- 3, Relative position of Asterix w.r.t. Earth
- 6, Relative velocity of Asterix w.r.t. Earth
- 9, LVLH to inertial frame rotation matrix of Asterix w.r.t. Earth

Warning, cannot reset dependentOrientationCalculator, incompatible object already exists

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

- 0, Single acceleration in inertial frame of type thrust , acting on Asterix
- 3, Relative position of Asterix w.r.t. Earth
- 6, Relative velocity of Asterix w.r.t. Earth
- 9, LVLH to inertial frame rotation matrix of Asterix w.r.t. Earth

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

- 0, Mach number of Apollo
- 1, Airspeed of Apollo
- 2, Density of Apollo
- 3, Single acceleration norm of type thrust , acting on Apollo

4, Body mass rate of Apollo

Warning, overriding existing angle of attack function in AerodynamicAngleCalculator

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Mach number of Apollo

1, Airspeed of Apollo

2, Density of Apollo

3, Single acceleration norm of type thrust , acting on Apollo

4, Body mass rate of Apollo

Warning, overriding existing angle of attack function in AerodynamicAngleCalculator

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Mach number of Apollo

1, Airspeed of Apollo

2, Density of Apollo

3, Single acceleration norm of type thrust , acting on Apollo

4, Body mass rate of Apollo

Warning, overriding existing angle of attack function in AerodynamicAngleCalculator

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Mach number of Apollo

1, Airspeed of Apollo

2, Density of Apollo

3, Single acceleration norm of type thrust , acting on Apollo

4, Body mass rate of Apollo

Warning, overriding existing angle of attack function in AerodynamicAngleCalculator

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Mach number of Apollo

1, Airspeed of Apollo

2, Density of Apollo

3, Single acceleration norm of type thrust , acting on Apollo

4, Body mass rateof Apollo

Warning, overriding existing angle of attack function in AerodynamicAngleCalculator

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Mach number of Apollo

1, Airspeed of Apollo

2, Density of Apollo

3, Single acceleration norm of type thrust , acting on Apollo

4, Body mass rateof Apollo

Warning, overriding existing angle of attack function in AerodynamicAngleCalculator

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Mach number of Apollo

1, Airspeed of Apollo

2, Density of Apollo

3, Single acceleration norm of type thrust , acting on Apollo

4, Body mass rateof Apollo

Warning, overriding existing angle of attack function in AerodynamicAngleCalculator

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Mach number of Apollo

1, Airspeed of Apollo

2, Density of Apollo

3, Single acceleration norm of type thrust , acting on Apollo

4, Body mass rateof Apollo

Dependent variables being saved, output vectors contain:

Vector entry, Vector contents

0, Single acceleration norm of type thrust , acting on Apollo

*** No errors detected

<end of output>

Test time = 18.19 sec

Test Passed.

"test_ThrustAcceleration" end time: Feb 17 12:30 W. Europe Standard Time

"test_ThrustAcceleration" time elapsed: 00:00:18

85/137 Testing: test_GroundStationState

85/137 Test: test_GroundStationState

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_GroundStationState.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Astrodynamics/GroundStations

"test_GroundStationState" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 2 test cases...

*** No errors detected

<end of output>

Test time = 2.31 sec

Test Passed.

"test_GroundStationState" end time: Feb 17 12:30 W. Europe Standard Time

"test_GroundStationState" time elapsed: 00:00:02

86/137 Testing: test_TimeTypes

86/137 Test: test_TimeTypes

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_TimeTypes.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Basics

"test_TimeTypes" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 4 test cases...

*** No errors detected

<end of output>

Test time = 1.85 sec

Test Passed.

"test_TimeTypes" end time: Feb 17 12:30 W. Europe Standard Time

"test_TimeTypes" time elapsed: 00:00:01

87/137 Testing: test_MatrixTextFileReader

87/137 Test: test_MatrixTextFileReader

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_MatrixTextFileReader.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/InOutOutput

"test_MatrixTextFileReader" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 1.86 sec

Test Passed.

"test_MatrixTextFileReader" end time: Feb 17 12:30 W. Europe Standard Time

"test_MatrixTextFileReader" time elapsed: 00:00:01

88/137 Testing: test_StreamFilters

88/137 Test: test_StreamFilters

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_StreamFilters.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/InOutOutput

"test_StreamFilters" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 8 test cases...

*** No errors detected

<end of output>

Test time = 1.84 sec

Test Passed.

"test_StreamFilters" end time: Feb 17 12:30 W. Europe Standard Time

"test_StreamFilters" time elapsed: 00:00:01

89/137 Testing: test_TwoLineElementsTextFileReader

89/137 Test: test_TwoLineElementsTextFileReader

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_TwoLineElementsTextFileReader.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/InOutOutput

"test_TwoLineElementsTextFileReader" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 2 test cases...

*** No errors detected

<end of output>

Test time = 2.01 sec

Test Passed.

"test_TwoLineElementsTextFileReader" end time: Feb 17 12:30 W. Europe Standard Time

"test_TwoLineElementsTextFileReader" time elapsed: 00:00:02

90/137 Testing: test_BasicInputOutput

90/137 Test: test_BasicInputOutput

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_BasicInputOutput.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/InputOutput

"test_BasicInputOutput" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 3 test cases...

*** No errors detected

<end of output>

Test time = 1.70 sec

Test Passed.

"test_BasicInputOutput" end time: Feb 17 12:30 W. Europe Standard Time

"test_BasicInputOutput" time elapsed: 00:00:01

91/137 Testing: test_ParsedDataVectorUtilities

91/137 Test: test_ParsedDataVectorUtilities

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_ParsedDataVectorUtilities.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/InputOutput

"test_ParsedDataVectorUtilities" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 3 test cases...

*** No errors detected

<end of output>

Test time = 1.64 sec

Test Passed.

"test_ParsedDataVectorUtilities" end time: Feb 17 12:30 W. Europe Standard Time

"test_ParsedDataVectorUtilities" time elapsed: 00:00:01

92/137 Testing: test_FieldValue

92/137 Test: test_FieldValue

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_FieldValue.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/InOutOutput

"test_FieldValue" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 5 test cases...

*** No errors detected

<end of output>

Test time = 1.71 sec

Test Passed.

"test_FieldValue" end time: Feb 17 12:30 W. Europe Standard Time

"test_FieldValue" time elapsed: 00:00:01

93/137 Testing: test_TextParser

93/137 Test: test_TextParser

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_TextParser.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/InOutOutput

"test_TextParser" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 2 test cases...

*** No errors detected

<end of output>

Test time = 1.86 sec

Test Passed.

"test_TextParser" end time: Feb 17 12:30 W. Europe Standard Time

"test_TextParser" time elapsed: 00:00:01

94/137 Testing: test_MissileDatcomReader

94/137 Test: test_MissileDatcomReader

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_MissileDatcomReader.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/InOutOutput

"test_MissileDatcomReader" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 1.85 sec

Test Passed.

"test_MissileDatcomReader" end time: Feb 17 12:30 W. Europe Standard Time

"test_MissileDatcomReader" time elapsed: 00:00:01

95/137 Testing: test_MissileDatcomData

95/137 Test: test_MissileDatcomData

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_MissileDatcomData.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/InOutOutput

"test_MissileDatcomData" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 2.30 sec

Test Passed.

"test_MissileDatcomData" end time: Feb 17 12:30 W. Europe Standard Time

"test_MissileDatcomData" time elapsed: 00:00:02

96/137 Testing: test_DictionaryInputSystem

96/137 Test: test_DictionaryInputSystem

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_DictionaryInputSystem.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/InputOutput

"test_DictionaryInputSystem" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 1.81 sec

Test Passed.

"test_DictionaryInputSystem" end time: Feb 17 12:30 W. Europe Standard Time

"test_DictionaryInputSystem" time elapsed: 00:00:01

97/137 Testing: test_Extractor

97/137 Test: test_Extractor

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_Extractor.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/InputOutput

"test_Extractor" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 3 test cases...

*** No errors detected

<end of output>

Test time = 1.78 sec

Test Passed.

"test_Extractor" end time: Feb 17 12:30 W. Europe Standard Time

"test_Extractor" time elapsed: 00:00:01

98/137 Testing: test_FixedWidthParser

98/137 Test: test_FixedWidthParser

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_FixedWidthParser.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/InputOutput

"test_FixedWidthParser" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 2 test cases...

*** No errors detected

<end of output>

Test time = 3.05 sec

Test Passed.

"test_FixedWidthParser" end time: Feb 17 12:30 W. Europe Standard Time

"test_FixedWidthParser" time elapsed: 00:00:03

99/137 Testing: test_SeparatedParser

99/137 Test: test_SeparatedParser

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_SeparatedParser.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/InOutOutput

"test_SeparatedParser" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 4 test cases...

*** No errors detected

<end of output>

Test time = 1.72 sec

Test Passed.

"test_SeparatedParser" end time: Feb 17 12:30 W. Europe Standard Time

"test_SeparatedParser" time elapsed: 00:00:01

100/137 Testing: test_LinearFieldTransform

100/137 Test: test_LinearFieldTransform

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_LinearFieldTransform.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/InOutOutput

"test_LinearFieldTransform" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 4 test cases...

*** No errors detected

<end of output>

Test time = 1.88 sec

Test Passed.

"test_LinearFieldTransform" end time: Feb 17 12:30 W. Europe Standard Time

"test_LinearFieldTransform" time elapsed: 00:00:01

101/137 Testing: test_SolarActivityData

101/137 Test: test_SolarActivityData

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_SolarActivityData.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/InputOutput

"test_SolarActivityData" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 2 test cases...

*** No errors detected

<end of output>

Test time = 4.76 sec

Test Passed.

"test_SolarActivityData" end time: Feb 17 12:30 W. Europe Standard Time

"test_SolarActivityData" time elapsed: 00:00:04

102/137 Testing: test_MultiArrayReader

102/137 Test: test_MultiArrayReader

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_MultiArrayReader.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/InputOutput

"test_MultiArrayReader" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 1.80 sec

Test Passed.

"test_MultiArrayReader" end time: Feb 17 12:30 W. Europe Standard Time

"test_MultiArrayReader" time elapsed: 00:00:01

103/137 Testing: test_AerodynamicCoefficientReader

103/137 Test: test_AerodynamicCoefficientReader

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_AerodynamicCoefficientReader.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/InputOutput

"test_AerodynamicCoefficientReader" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 1.85 sec

Test Passed.

"test_AerodynamicCoefficientReader" end time: Feb 17 12:30 W. Europe Standard Time

"test_AerodynamicCoefficientReader" time elapsed: 00:00:01

104/137 Testing: test_MathematicalConstants

104/137 Test: test_MathematicalConstants

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_MathematicalConstants.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/BasicMathematics

"test_MathematicalConstants" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 5 test cases...

*** No errors detected

<end of output>

Test time = 2.04 sec

Test Passed.

"test_MathematicalConstants" end time: Feb 17 12:30 W. Europe Standard Time

"test_MathematicalConstants" time elapsed: 00:00:02

105/137 Testing: test_LinearAlgebra

105/137 Test: test_LinearAlgebra

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_LinearAlgebra.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/BasicMathematics

"test_LinearAlgebra" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 1.80 sec

Test Passed.

"test_LinearAlgebra" end time: Feb 17 12:30 W. Europe Standard Time

"test_LinearAlgebra" time elapsed: 00:00:01

106/137 Testing: test_CoordinateConversions

106/137 Test: test_CoordinateConversions

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_CoordinateConversions.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/BasicMathematics

"test_CoordinateConversions" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 10 test cases...

*** No errors detected

<end of output>

Test time = 1.62 sec

Test Passed.

"test_CoordinateConversions" end time: Feb 17 12:30 W. Europe Standard Time

"test_CoordinateConversions" time elapsed: 00:00:01

107/137 Testing: test_NearestNeighbourSearch

107/137 Test: test_NearestNeighbourSearch

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_NearestNeighbourSearch.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/BasicMathematics

"test_NearestNeighbourSearch" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 1.68 sec

Test Passed.

"test_NearestNeighbourSearch" end time: Feb 17 12:30 W. Europe Standard Time

"test_NearestNeighbourSearch" time elapsed: 00:00:01

108/137 Testing: test_NumericalDerivative

108/137 Test: test_NumericalDerivative

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_NumericalDerivative.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/BasicMathematics

"test_NumericalDerivative" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 1.75 sec

Test Passed.

"test_NumericalDerivative" end time: Feb 17 12:30 W. Europe Standard Time

"test_NumericalDerivative" time elapsed: 00:00:01

109/137 Testing: test_LegendrePolynomials

109/137 Test: test_LegendrePolynomials

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_LegendrePolynomials.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/BasicMathematics

"test_LegendrePolynomials" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 4 test cases...

*** No errors detected

<end of output>

Test time = 1.75 sec

Test Passed.

"test_LegendrePolynomials" end time: Feb 17 12:30 W. Europe Standard Time

"test_LegendrePolynomials" time elapsed: 00:00:01

110/137 Testing: test_SphericalHarmonics

110/137 Test: test_SphericalHarmonics

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_SphericalHarmonics.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/BasicMathematics

"test_SphericalHarmonics" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 1.66 sec

Test Passed.

"test_SphericalHarmonics" end time: Feb 17 12:30 W. Europe Standard Time

"test_SphericalHarmonics" time elapsed: 00:00:01

111/137 Testing: test_RotationAboutArbitraryAxis

111/137 Test: test_RotationAboutArbitraryAxis

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_RotationAboutArbitraryAxis.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/BasicMathematics

"test_RotationAboutArbitraryAxis" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 5 test cases...

*** No errors detected

<end of output>

Test time = 1.91 sec

Test Passed.

"test_RotationAboutArbitraryAxis" end time: Feb 17 12:30 W. Europe Standard Time

"test_RotationAboutArbitraryAxis" time elapsed: 00:00:01

112/137 Testing: test_LawgsSurfaceGeometry

112/137 Test: test_LawgsSurfaceGeometry

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_LawgsSurfaceGeometry.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/GeometricShapes

"test_LawgsSurfaceGeometry" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 1.74 sec

Test Passed.

"test_LawgsSurfaceGeometry" end time: Feb 17 12:30 W. Europe Standard Time

"test_LawgsSurfaceGeometry" time elapsed: 00:00:01

113/137 Testing: test_CubicSplineInterpolator

113/137 Test: test_CubicSplineInterpolator

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_CubicSplineInterpolator.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/Interpolators

"test_CubicSplineInterpolator" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 4 test cases...

*** No errors detected

<end of output>

Test time = 2.09 sec

Test Passed.

"test_CubicSplineInterpolator" end time: Feb 17 12:30 W. Europe Standard Time

"test_CubicSplineInterpolator" time elapsed: 00:00:02

114/137 Testing: test_HermiteCubicSplineInterpolator

114/137 Test: test_HermiteCubicSplineInterpolator

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_HermiteCubicSplineInterpolator.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/Interpolators

"test_HermiteCubicSplineInterpolator" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 2 test cases...

*** No errors detected

<end of output>

Test time = 2.03 sec

Test Passed.

"test_HermiteCubicSplineInterpolator" end time: Feb 17 12:30 W. Europe Standard Time

"test_HermiteCubicSplineInterpolator" time elapsed: 00:00:02

115/137 Testing: test_LinearInterpolator

115/137 Test: test_LinearInterpolator

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_LinearInterpolator.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/Interpolators

"test_LinearInterpolator" start time: Feb 17 12:30 W. Europe Standard Time

Output:

Running 3 test cases...

*** No errors detected

<end of output>

Test time = 2.23 sec

Test Passed.

"test_LinearInterpolator" end time: Feb 17 12:31 W. Europe Standard Time

"test_LinearInterpolator" time elapsed: 00:00:02

116/137 Testing: test_MultiLinearInterpolator

116/137 Test: test_MultiLinearInterpolator

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_MultiLinearInterpolator.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/Interpolators

"test_MultiLinearInterpolator" start time: Feb 17 12:31 W. Europe Standard Time

Output:

Running 2 test cases...

*** No errors detected

<end of output>

Test time = 1.65 sec

Test Passed.

"test_MultiLinearInterpolator" end time: Feb 17 12:31 W. Europe Standard Time

"test_MultiLinearInterpolator" time elapsed: 00:00:01

117/137 Testing: test_LagrangeInterpolator

117/137 Test: test_LagrangeInterpolator

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_LagrangeInterpolator.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/Interpolators

"test_LagrangeInterpolator" start time: Feb 17 12:31 W. Europe Standard Time

Output:

Running 3 test cases...

*** No errors detected

<end of output>

Test time = 2.00 sec

Test Passed.

"test_LagrangeInterpolator" end time: Feb 17 12:31 W. Europe Standard Time

"test_LagrangeInterpolator" time elapsed: 00:00:01

118/137 Testing: test_EulerIntegrator

118/137 Test: test_EulerIntegrator

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_EulerIntegrator.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/NumericalIntegrators

"test_EulerIntegrator" start time: Feb 17 12:31 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 1.85 sec

Test Passed.

"test_EulerIntegrator" end time: Feb 17 12:31 W. Europe Standard Time

"test_EulerIntegrator" time elapsed: 00:00:01

119/137 Testing: test_NumericalIntegrator

119/137 Test: test_NumericalIntegrator

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_NumericalIntegrator.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/NumericalIntegrators

"test_NumericalIntegrator" start time: Feb 17 12:31 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 1.80 sec

Test Passed.

"test_NumericalIntegrator" end time: Feb 17 12:31 W. Europe Standard Time

"test_NumericalIntegrator" time elapsed: 00:00:01

120/137 Testing: test_RungeKutta4Integrator

120/137 Test: test_RungeKutta4Integrator

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_RungeKutta4Integrator.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/NumericalIntegrators

"test_RungeKutta4Integrator" start time: Feb 17 12:31 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 1.86 sec

Test Passed.

"test_RungeKutta4Integrator" end time: Feb 17 12:31 W. Europe Standard Time

"test_RungeKutta4Integrator" time elapsed: 00:00:01

121/137 Testing: test_RungeKuttaVariableStepSizeIntegrator

121/137 Test: test_RungeKuttaVariableStepSizeIntegrator

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_RungeKuttaVariableStepSizeIntegrator.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/NumericalIntegrators

"test_RungeKuttaVariableStepSizeIntegrator" start time: Feb 17 12:31 W. Europe Standard Time

Output:

Running 4 test cases...

*** No errors detected

<end of output>

Test time = 1.81 sec

Test Passed.

"test_RungeKuttaVariableStepSizeIntegrator" end time: Feb 17 12:31 W. Europe Standard Time

"test_RungeKuttaVariableStepSizeIntegrator" time elapsed: 00:00:01

122/137 Testing: test_RungeKuttaCoefficients

122/137 Test: test_RungeKuttaCoefficients

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_RungeKuttaCoefficients.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/NumericalIntegrators

"test_RungeKuttaCoefficients" start time: Feb 17 12:31 W. Europe Standard Time

Output:

Running 3 test cases...

*** No errors detected

<end of output>

Test time = 1.66 sec

Test Passed.

"test_RungeKuttaCoefficients" end time: Feb 17 12:31 W. Europe Standard Time

"test_RungeKuttaCoefficients" time elapsed: 00:00:01

123/137 Testing: test_RungeKuttaFehlberg45Integrator

123/137 Test: test_RungeKuttaFehlberg45Integrator

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_RungeKuttaFehlberg45Integrator.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/NumericalIntegrators

"test_RungeKuttaFehlberg45Integrator" start time: Feb 17 12:31 W. Europe Standard Time

Output:

Running 2 test cases...

*** No errors detected

<end of output>

Test time = 1.87 sec

Test Passed.

"test_RungeKuttaFehlberg45Integrator" end time: Feb 17 12:31 W. Europe Standard Time

"test_RungeKuttaFehlberg45Integrator" time elapsed: 00:00:01

124/137 Testing: test_RungeKuttaFehlberg56Integrator

124/137 Test: test_RungeKuttaFehlberg56Integrator

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_RungeKuttaFehlberg56Integrator.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/NumericalIntegrators

"test_RungeKuttaFehlberg56Integrator" start time: Feb 17 12:31 W. Europe Standard Time

Output:

Running 4 test cases...

*** No errors detected

<end of output>

Test time = 1.76 sec

Test Passed.

"test_RungeKuttaFehlberg56Integrator" end time: Feb 17 12:31 W. Europe Standard Time

"test_RungeKuttaFehlberg56Integrator" time elapsed: 00:00:01

125/137 Testing: test_RungeKuttaFehlberg78Integrator

125/137 Test: test_RungeKuttaFehlberg78Integrator

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_RungeKuttaFehlberg78Integrator.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/NumericalIntegrators

"test_RungeKuttaFehlberg78Integrator" start time: Feb 17 12:31 W. Europe Standard Time

Output:

Running 2 test cases...

*** No errors detected

<end of output>

Test time = 2.00 sec

Test Passed.

"test_RungeKuttaFehlberg78Integrator" end time: Feb 17 12:31 W. Europe Standard Time

"test_RungeKuttaFehlberg78Integrator" time elapsed: 00:00:02

126/137 Testing: test_RungeKutta87DormandPrinceIntegrator

126/137 Test: test_RungeKutta87DormandPrinceIntegrator

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_RungeKutta87DormandPrinceIntegrator.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/NumericalIntegrators

"test_RungeKutta87DormandPrinceIntegrator" start time: Feb 17 12:31 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 1.92 sec

Test Passed.

"test_RungeKutta87DormandPrinceIntegrator" end time: Feb 17 12:31 W. Europe Standard Time

"test_RungeKutta87DormandPrinceIntegrator" time elapsed: 00:00:01

127/137 Testing: test_RootFinders

127/137 Test: test_RootFinders

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_RootFinders.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/RootFinders

"test_RootFinders" start time: Feb 17 12:31 W. Europe Standard Time

Output:

Running 19 test cases...

*** No errors detected

<end of output>

Test time = 4.87 sec

Test Passed.

"test_RootFinders" end time: Feb 17 12:31 W. Europe Standard Time

"test_RootFinders" time elapsed: 00:00:04

128/137 Testing: test_SimpleLinearRegression

128/137 Test: test_SimpleLinearRegression

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_SimpleLinearRegression.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/Statistics

"test_SimpleLinearRegression" start time: Feb 17 12:31 W. Europe Standard Time

Output:

Running 3 test cases...

*** No errors detected

<end of output>

Test time = 1.74 sec

Test Passed.

"test_SimpleLinearRegression" end time: Feb 17 12:31 W. Europe Standard Time

"test_SimpleLinearRegression" time elapsed: 00:00:01

129/137 Testing: test_BasicStatistics

129/137 Test: test_BasicStatistics

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_BasicStatistics.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/Statistics

"test_BasicStatistics" start time: Feb 17 12:31 W. Europe Standard Time

Output:

Running 2 test cases...

*** No errors detected

<end of output>

Test time = 1.95 sec

Test Passed.

"test_BasicStatistics" end time: Feb 17 12:31 W. Europe Standard Time

"test_BasicStatistics" time elapsed: 00:00:01

130/137 Testing: test_MultiVariateProbabilityDistributions

130/137 Test: test_MultiVariateProbabilityDistributions

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_MultiVariateProbabilityDistributions.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/Statistics

"test_MultiVariateProbabilityDistributions" start time: Feb 17 12:31 W. Europe Standard Time

Output:

Running 2 test cases...

*** No errors detected

<end of output>

Test time = 2.20 sec

Test Passed.

"test_MultiVariateProbabilityDistributions" end time: Feb 17 12:31 W. Europe Standard Time

"test_MultiVariateProbabilityDistributions" time elapsed: 00:00:02

131/137 Testing: test_BoostDistributions

131/137 Test: test_BoostDistributions

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_BoostDistributions.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/Statistics

"test_BoostDistributions" start time: Feb 17 12:31 W. Europe Standard Time

Output:

Running 1 test case...

*** No errors detected

<end of output>

Test time = 5.03 sec

Test Passed.

"test_BoostDistributions" end time: Feb 17 12:31 W. Europe Standard Time

"test_BoostDistributions" time elapsed: 00:00:05

132/137 Testing: test_KernelDensityDistribution

132/137 Test: test_KernelDensityDistribution

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_KernelDensityDistribution.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/Statistics

"test_KernelDensityDistribution" start time: Feb 17 12:31 W. Europe Standard Time

Output:

Running 7 test cases...

*** No errors detected

<end of output>

Test time = 8.62 sec

Test Passed.

"test_KernelDensityDistribution" end time: Feb 17 12:31 W. Europe Standard Time

"test_KernelDensityDistribution" time elapsed: 00:00:08

133/137 Testing: test_RandomSampling

133/137 Test: test_RandomSampling

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_RandomSampling.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/Statistics

"test_RandomSampling" start time: Feb 17 12:31 W. Europe Standard Time

Output:

Running 2 test cases...

*** No errors detected

<end of output>

Test time = 5.90 sec

Test Passed.

"test_RandomSampling" end time: Feb 17 12:31 W. Europe Standard Time

"test_RandomSampling" time elapsed: 00:00:05

134/137 Testing: test_TrapezoidalIntegrator

134/137 Test: test_TrapezoidalIntegrator

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_TrapezoidalIntegrator.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/Mathematics/NumericalQuadrature

"test_TrapezoidalIntegrator" start time: Feb 17 12:31 W. Europe Standard Time

Output:

Running 2 test cases...

*** No errors detected

<end of output>

Test time = 2.32 sec

Test Passed.

"test_TrapezoidalIntegrator" end time: Feb 17 12:31 W. Europe Standard Time

"test_TrapezoidalIntegrator" time elapsed: 00:00:02

135/137 Testing: test_SpiceInterface

135/137 Test: test_SpiceInterface

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_SpiceInterface.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/External/SpiceInterface

"test_SpiceInterface" start time: Feb 17 12:31 W. Europe Standard Time

Output:

Running 7 test cases...

*** No errors detected

<end of output>

Test time = 2.68 sec

Test Passed.

"test_SpiceInterface" end time: Feb 17 12:31 W. Europe Standard Time

"test_SpiceInterface" time elapsed: 00:00:02

136/137 Testing: test_EnvironmentSetup

136/137 Test: test_EnvironmentSetup

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_EnvironmentSetup.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/SimulationSetup

"test_EnvironmentSetup" start time: Feb 17 12:31 W. Europe Standard Time

Output:

<end of output>

Test time = 0.00 sec

Test Failed.

"test_EnvironmentSetup" end time: Feb 17 12:31 W. Europe Standard Time

"test_EnvironmentSetup" time elapsed: 00:00:00

137/137 Testing: test_AccelerationModelSetup

137/137 Test: test_AccelerationModelSetup

Command: "C:/tudatBundle/tudat/bin/unit_tests/test_AccelerationModelSetup.exe"

Directory: C:/build-tudatBundle-Desktop-Default/tudat/Tudat/SimulationSetup

"test_AccelerationModelSetup" start time: Feb 17 12:31 W. Europe Standard Time

Output:

<end of output>

Test time = 0.00 sec

Test Failed.

"test_AccelerationModelSetup" end time: Feb 17 12:31 W. Europe Standard Time

"test_AccelerationModelSetup" time elapsed: 00:00:00

End testing: Feb 17 12:31 W. Europe Standard Time